

GENERAL MOTORS CORP INDIANAPOLIS IN DETROIT DIESEL A--ETC F/G 21/5
ALLISON PD370-42 ADVANCED TURBOPROP ENGINE.(U)
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20. ABSTRACT (Continue on reverse side if necessary and identify by block number) This study developed data on Detroit Diesel Allison (DDA) common core derivative engines for use in Maritime Patrol Aircraft (MPA) concept formulation studies. The study included the screening of potential DDA turboprop/turboshift engines and the preparation of technical and planning information on three of the most promising engine candidates plus an all new engine. Screening of DDA derivative candidates was performed utilizing an analytical MPA model using synthesized mission profiles to rank the candidates in terms		

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of fuel consumption, weight, cost and complexity. The three turboprop engines selected for further study were as follows: a derivative of the unity size T701-AD-700 shaft power engine with rematched turbine (PD 370-37), an advanced T701 turboprop derivative with 25:1 overall pressure ratio and a scaled ATEGG demonstrated compressor (PD 370-40), an advanced T701 turboprop derivative with 17.7:1 overall pressure ratio and a scaled ATEGG demonstrated compressor (PD 370-41). Data is also presented on a new advanced turboprop engine with 30:1 overall pressure ratio which incorporates compressor, combustor, turbine, and cooling technology now under development and demonstration at DDA. The documentation consists of six separate reports prepared in the following manner. One report summarizes the engine screening analysis and describes the approach to, and the conclusions of the study. A separate report for each of the three derivative engines and for the new turboprop present estimates of performance, weight, and dimensional data. The engineering budgetary estimates of the development, acquisition, and service costs for each of the four engines are presented in a separate report.

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REVISIONS

<u>Letter</u>	<u>Page</u>	<u>Revision</u>
A	4	All weights
A	6	All weights
A	10	Additional matrix points at 0 and 25,000 feet
A	13	Additional performance
A	18	Additional performance
A	23	Additional performance
A	28	Additional performance



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I. INTRODUCTION

This report presents estimates of performance, weight, and dimensional data for the PD 370-42 turboprop engine. The PD 370-42 is an advanced turboprop engine with 30:1 overall pressure ratio. It incorporates compressor, combustor, turbine, and cooling techniques now under development and demonstration at DDA. The technologies represent the highest levels that can be offered for IOC's circa 1990's. The engine is in the 9,000 to 10,000 SHP class, but scaling data is included to provide for studies down to 6,000 and up to 12,000 SHP. The data is submitted for use in preliminary design type studies in the evaluation of turboprop systems.

The reduction gearbox for speed reduction to the prop-fan is a new simplified design, compared to the DDA T56 series of gearboxes. The new design is based upon a study into the reliability and maintenance cost history of past turboprop systems, and follows the recommendations of that study for a gearbox with high reliability, easy maintainability, and low maintenance costs.



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II. ENGINE DESCRIPTION

The Model PD 370-42 is an axial flow engine, having a single spool core and a free power turbine connected by shafting, and supporting structure to an offset reduction gear assembly. The general arrangement and external features of the engine are shown in Figure II-1, with principle physical characteristics listed in Table II-I. Output speed of the engine is constant at 12,860 RPM. The reduction gearbox shown in Figure II-1 has an overall gear ratio of 9.52:1, providing a propfan speed of 1351 rpm at 12,860 engine rpm. Parametric weight data is shown in Section III so that other propfan rotational speeds, and gear ratios can be analyzed. An aircraft accessory drive pad is provided on the back of the gearbox to drive an aircraft mounted accessory drive box. Power available at this pad is 500 HP at 8000 rpm. The primary engine mounts are on the gearbox with a hang mount at the rear of the engine. Engine accessories are driven by a bevel drive from the high pressure spool. The control system is integral with the propfan and is digital electronic. The oil system is integral to the engine and also supplies the propfan and reduction gearbox, but is separately filtered and monitored to isolate fault detection in each of these major modules. Engine torque is measured hydraulically from the gear thrust of the power train idler gears in the reduction gearbox.

The gearbox is shown offset, based upon DDA's experience with large turboprop engines. It is offset-up to be consistent with current studies showing a preference to under-the-wing engine mounting. It can also be supplied in the offset-down position.

Performance ratings, sea level static, are listed in Table II-II.

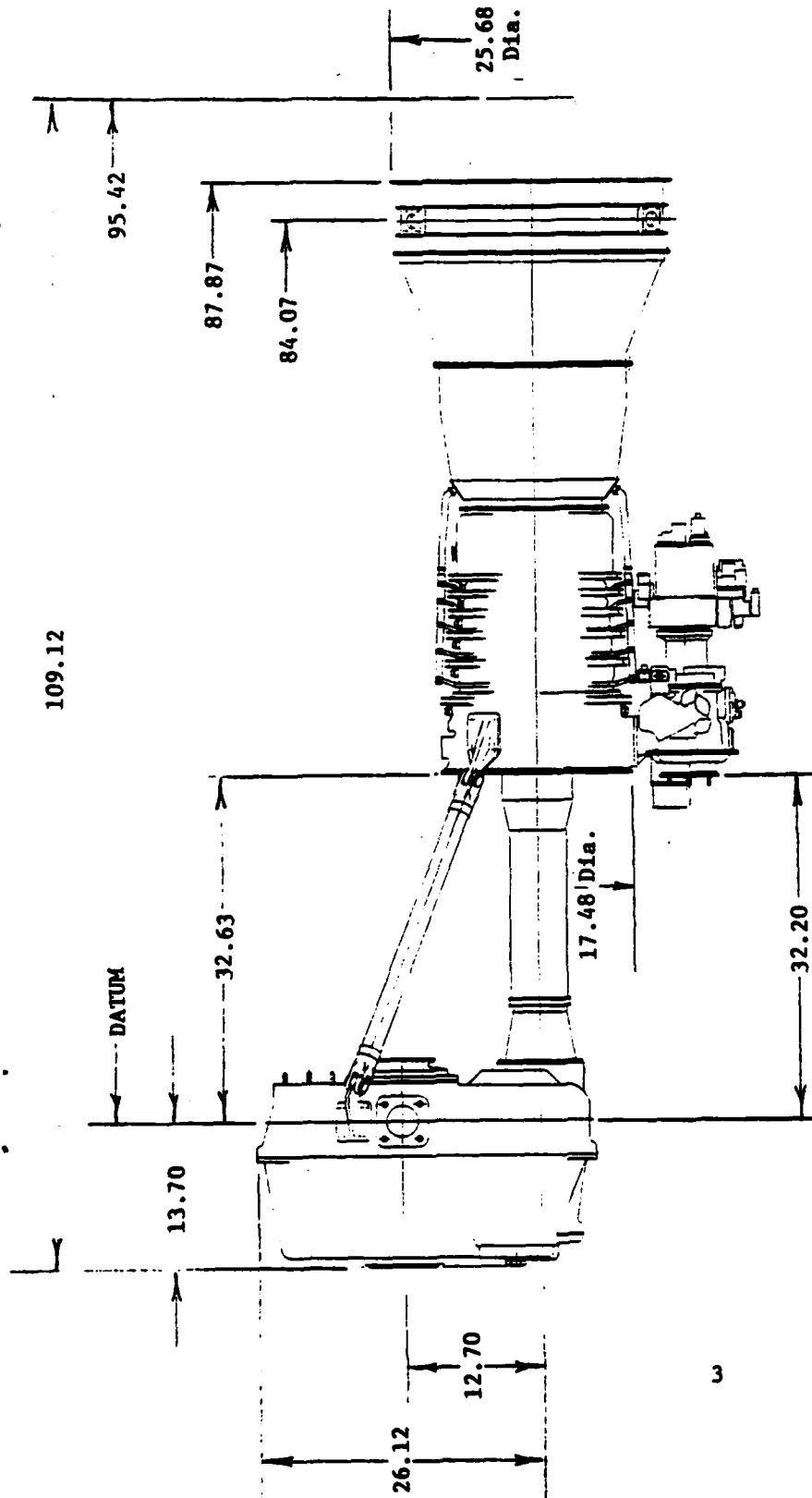
For preliminary design studies, the PD 370-42 engine configuration can be scaled to other power ratings. Scaling information is included for dimensions, weight, and performance.



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Figure 11-1. PD370-42 General Arrangement.



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TABLE II-I

PD 370-42 PHYSICAL CHARACTERISTICS

(Includes Gearbox)

Length (in)	109.12
Max. Engine Diameter (in)	25.68
Max. Gearbox Offset, upward (in)	26.12
Dry Weight, lbs	
Engine	886 889
Gearbox, including interconnecting struts and shaft	585 468
Total	1471 1367

For scaling dimensions the following formulae may be used for SHP's up to 12,000 and down to 6,000, and for other reduction gear ratios than 9.52:1.

Engine

$$\text{Axial dimensions} = \text{Base dim.} \times \left(\frac{\text{SHP}}{9,610} \right)^{0.4}$$

$$\text{Diameters} = \text{Base dia.} \times \left(\frac{\text{SHP}}{9,610} \right)^{0.5}$$

Reduction Gearbox

$$\text{Dimensions} = \text{Base dim.} \times \left(\frac{\text{SHP}}{9,610} \right)^{0.5} \times \left(\frac{\text{GR}}{9.52} \right)^{0.33}$$

Shaft length remains unchanged.



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TABLE II-II

PD370-42 PERFORMANCE SUMMARY

Sea Level, 0 kts

	<u>Standard Day</u>			<u>Hot Day, 89.8°F</u>		
	SHP	SFC	F _N	SHP	SFC	F _N
Take-Off	9,610	0.348	827	7,924	0.363	645
Max. Continuous	7,302	0.363	604	5,989	0.381	474



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III. WEIGHTS

The weight of the basic 9,610 SHP engine, gearbox, and the interconnecting struts and shaft are given in Table III-I. The gearbox weight is based upon a gear ratio of 9.52:1 which provides a propfan speed of 1351 rpm.

TABLE III-I

PD 370-42 WEIGHTS

	<u>Dry</u>	<u>Wet</u> [*]	<u>Installed</u>
Basic Engine, lbs	886	907	907
Gearbox, lbs	557	597	597
Interconnecting Struts and Shaft, lbs	28	28	28
Total, lbs	1471	1532	1532

* Includes total amount of oil required for engine and gearbox operation.

For scaling weights to engine sizes up to 12,000 and down to 6,000 SHP, and for other reduction gear ratios, the following formulae may be used:

$$\text{Engine weight} = 886 \times \left(\frac{\text{HP}}{9,610} \right)^{1.01}$$

$$\text{Gearbox weight} = 557 \times \left(\frac{\text{HP}}{9,610} \right)^{1.5} \times \left(\frac{\text{GR}}{9.52} \right)^{0.4}$$

Interconnecting strut and shaft weight = 5.1% of dry gearbox weight



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IV. STEADY STATE PERFORMANCE

Steady state performance data is tabulated in this section for all points shown in Figure IV-1. Basic engine data is shown for the following assumptions:

- o Uninstalled engine
- o ICAO standard atmosphere except for takeoff which in addition includes an ambient temperature of 89.8°F at standard atmosphere
- o 100% inlet recovery
- o Zero accessory horsepower extraction
- o Zero customer bleed extraction
- o Zero losses due to reduction gear
- o Fuel heating value - 18,400 Btu/lb
- o Estimated average engine performance - No SHP or fuel flow guarantee factors

Sensitivity data is provided for each point so that bleed and duct losses may be estimated as required.

Nomenclature

Nomenclature used in the tabulation of performance is as follows:

MACH	Mach number
SHP	Shaft horsepower
SFC	Specific fuel consumption, lbs/hr/hp
WF	Engine fuel flow, lbs/hr
FN	Net jet thrust, lbs (jet gross thrust - ram drag)
ESHP	Equivalent shaft horsepower (energy in jet stream converted ideally to horsepower and added to SHP)
WCIN	Total inlet corrected airflow, $W\sqrt{\theta_1}/\delta_1$
	where: θ_1 = Engine inlet total temp, °R

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$$\delta_1 = \frac{\text{Engine inlet total pressure, psi}}{14.696}$$

TNOZ Jet nozzle total temperature, °R
 PNOZ Jet nozzle total pressure, psi
 RC Compressor pressure ratio
 BOT Burner outlet temperature, °R
 NO Point number

Sensitivity Data

Bleed:

$$\text{SHP, with bleed} = \text{SHP, no bleed} - (\text{DEL SHP})(\% \text{ bleed})$$

$$\text{WF, with bleed} = \text{WF, no bleed} - (\text{DEL WF})(\% \text{ bleed})$$

$$\text{FN, with bleed} = \text{FN, no bleed} - (\text{DEL FN})(\% \text{ bleed})$$

Inlet Recovery:

$$\eta_R = \frac{\text{Total pressure actual}}{\text{Total pressure ideal}}$$

$$\text{SHP, with recovery} = \text{SHP, ideal recovery} - (\text{DEL SHP})(1 - \eta_R)(100)$$

$$\text{WF, with recovery} = \text{WF, ideal recovery} (\eta_R)$$

$$\text{FN, with recovery} = \text{FN, ideal recovery} - (\text{DEL FN})(1 - \eta_R)(100)$$

Jet Nozzle Duct Loss:

To estimate thrust loss due to additional duct loss prior to the jet nozzle, use the following equation:

$$\text{FN, with loss} = \text{FN, without loss} - \text{FN, without loss} (K) \left(\frac{\Delta P}{P} \right)$$

where,

o K is obtained for each point from sensitivity data

$$o \frac{\Delta P}{P} = \frac{\text{PTOT, no loss} - \text{PTOT, total loss}}{\text{PTOT, no loss}}$$



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Reduction Gear Loss:

Reduction gear is 99 percent efficient.

Accessory Drive Losses:

Accessory drive power extraction is directly from the accessory drive pad on the reduction gearbox. Reduce SHP to prop-fan by amount of accessory power extraction at each point.

Scale Effect on SHP and SFC:

The engine performance is scaleable for purposes of studying other engine sizes. However, component performance will vary, dependent upon the amount of scale. A correction to scale factor for SHP and SFC was formulated for component performance changes and the effects on SHP, SFC, and FN are as follows:

$$\text{Scaled SHP} = \text{Unscaled SHP} \times \left(1 - \frac{A}{100}\right) \times \text{Scale Factor}$$

$$\text{Scaled FN} = \text{Unscaled FN} \times \text{Scale Factor}$$

$$\text{Scaled SFC} = \text{Unscaled SFC} \times \left(1 + \frac{B}{100}\right)$$

where:

$$\text{Scale Factor} = \frac{\text{Desired Rating}}{\text{Unity Rating}} \quad (\text{limited from 0.6 to 1.25})$$

A = Sizing effect on SHP

$$= 11.94 \times \text{Scale factor} - 3.80 \times (\text{Scale factor})^2 - 8.14$$

B = Sizing effect on SFC

$$= 2.98 \times (\text{Scale factor})^2 - 9.46 \times \text{Scale factor} + 6.48$$

Nozzle Throat Area

The effective nozzle throat area is constant for all conditions at 263.9 in².



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Standard and Hot Day; Takeoff and Maximum Continuous

Altitude (Ft x 10 ⁻³)	MACH Number			
	0	.1	.2	.3
0	X	X	X	X

Standard Day; Maximum Climb, Maximum Continuous and Part Power to Idle

Altitude (Ft x 10 ⁻³)	MACH Number							
	.2	.3	.4	.5	.6	.7	.75	.8
0	X	X	X	X	X	X	X	X
5	X	X	X	X	X			
10	X	X	X	X	X	X	X	X
15		X	X	X	X	X	X	X
20			X	X	X	X	X	X
25			X	X	X	X	X	X
30				X	X	X	X	X
35				X	X	X	X	X
40				X	X	X	X	X
45				X	X	X	X	X

Figure IV-1. Matrix of flight conditions for performance data

DETROIT DIESEL ALLISON DIVISION				GENERAL MOTORS CORPORATION			
ZERO BLEED				ZERO POWER EXTRACTION			
POWER	MACH	SHP	SFC	WF	FN	ESHP	NCIN
TO	0.0	9810.	0.348	3748.	827.0	10201.	34.0
M.C.	0.10	7308.	0.362	3258.	903.3	7668.	29.6
TO	0.10	5946.	0.348	3258.	711.3	10241.	33.9
M.C.	0.10	7335.	0.362	3258.	503.1	7699.	29.6
TO	0.20	5743.	0.348	3258.	604.0	10350.	33.5
M.C.	0.20	7428.	0.360	3258.	401.0	7799.	29.3
TO	0.30	9901.	0.344	3258.	498.7	10527.	34.8
M.C.	0.30	7587.	0.357	2716.	313.1	17973.	28.8

T AMBIENT = 89.8°

DETROIT DIESEL ALLISON DIVISION				GENERAL MOTORS CORPORATION			
ZERO BLEED				ZERO POWER EXTRACTION			
POWER	MACH	SHP	SFC	WF	FN	ESHP	NCIN
TO	0.0	9810.	0.348	3748.	827.0	10201.	34.0
M.C.	0.10	7308.	0.362	3258.	903.3	7668.	29.6
TO	0.10	5946.	0.348	3258.	711.3	10241.	33.9
M.C.	0.10	7335.	0.362	3258.	503.1	7699.	29.6
TO	0.20	5743.	0.348	3258.	604.0	10350.	33.5
M.C.	0.20	7428.	0.360	3258.	401.0	7799.	29.3
TO	0.30	9901.	0.344	3258.	498.7	10527.	34.8
M.C.	0.30	7587.	0.357	2716.	313.1	17973.	28.8

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SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

0 FEET ALTITUDE

POWER	MACH	DEL SHP	INLET DEL FR	DEL SHP	PER DEL HP	BLEED DEL FR	DEL FR	K	NO
TO	0.0	13.4	17.3	193.3	3	43.8	21.2	4.40	0001
M.C.	0.0	13.4	17.3	193.3	3	43.8	21.2	4.40	0002
M.C.	0.10	13.4	17.3	193.3	3	43.8	21.2	4.40	0003
TO	0.20	13.4	17.3	193.3	3	43.8	21.2	4.40	0004
M.C.	0.30	13.4	17.3	193.3	3	43.8	21.2	4.40	0005
TO	0.30	13.4	17.3	193.3	3	43.8	21.2	4.40	0006
M.C.	0.30	13.4	17.3	193.3	3	43.8	21.2	4.40	0007
									0008

T AMBIENT = 89.8°F

0 FEET ALTITUDE

POWER	MACH	DEL SHP	INLET DEL FR	DEL SHP	PER DEL HP	BLEED DEL FR	DEL FR	K	NO
TO	0.0	15.7	18.0	193.3	3	43.8	21.2	4.40	0009
M.C.	0.0	15.7	18.0	193.3	3	43.8	21.2	4.40	0010
M.C.	0.10	15.7	18.0	193.3	3	43.8	21.2	4.40	0011
TO	0.20	15.7	18.0	193.3	3	43.8	21.2	4.40	0012
M.C.	0.30	15.7	18.0	193.3	3	43.8	21.2	4.40	0013
TO	0.30	15.7	18.0	193.3	3	43.8	21.2	4.40	0014
M.C.	0.30	15.7	18.0	193.3	3	43.8	21.2	4.40	0015
									0016

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ZERO POWER EXTRACTION

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100 PERCENT RECOVERY

STD DAY

0 FEET ALTITUDE

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ZERO BLEED

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ZERO POWER EXTRACTION

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PD370-42 TURBOPROP
100 PERCENT RECOVERY
STD DAY

DETROIT DIESEL ALLISON DIVISION
ZERO BLEED

[illegible]

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100 PERCENT RECOVERY
STD DAY

DETROIT DIESEL ALLISON DIVISION
ZERO BLEED

POWER	MACH	SHP	SFC	WF	FN	ESHP	WCIN	TNOZ	PN0Z	RC	NO
CLIMB M.C.	40	5901	2229	92	555	257	60	4	11	1	00
	40	5593	2229	97	555	257	60	4	11	1	00
	40	3692	3337	1252	555	257	60	4	11	1	00
	40	1872	3337	735	555	257	60	4	11	1	00
	40	1226	3337	536	555	257	60	4	11	1	00
CLIMB M.C.	40	6161	3337	1862	555	257	60	4	11	1	00
	40	5507	3337	1862	555	257	60	4	11	1	00
	40	3815	3337	1862	555	257	60	4	11	1	00
	40	1295	3337	1862	555	257	60	4	11	1	00
	40	1295	3337	1862	555	257	60	4	11	1	00
CLIMB M.C.	40	6036	3337	2041	555	257	60	4	11	1	00
	40	5507	3337	2041	555	257	60	4	11	1	00
	40	3970	3337	2041	555	257	60	4	11	1	00
	40	2929	3337	2041	555	257	60	4	11	1	00
CLIMB M.C.	40	1374	3337	1561	555	257	60	4	11	1	00
	40	6879	3337	1561	555	257	60	4	11	1	00
	40	6244	3337	1561	555	257	60	4	11	1	00
	40	3086	3337	1561	555	257	60	4	11	1	00
	40	1766	3337	1561	555	257	60	4	11	1	00
CLIMB M.C.	40	6770	3337	2192	555	257	60	4	11	1	00
	40	5507	3337	2192	555	257	60	4	11	1	00
	40	3275	3337	2192	555	257	60	4	11	1	00
	40	1766	3337	2192	555	257	60	4	11	1	00
CLIMB M.C.	40	7198	3337	1561	555	257	60	4	11	1	00
	40	6571	3337	1561	555	257	60	4	11	1	00
	40	3275	3337	1561	555	257	60	4	11	1	00
	40	1766	3337	1561	555	257	60	4	11	1	00

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PD370-42 TURBOPROP

100 PERCENT RECOVERY

STD DAY

25000 FEET ALTITUDE

GENERAL MOTORS CORPORATION

ZERO POWER EXTRACTION

POWER

UNTER
M.C.

J.M. INTER.

**INTER
M.C.**

**INTER
H.C.**

**INTER
H.C.**

INTER
M.C.

MACH

Q.40

DHS

35

134

FK

dHSE

N I J N

ZNQZ

ZONd

RC

10

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ZERO POWER EXTRACTION

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100 PERCENT RECOVERY
STD DAY

DETROIT DIESEL ALLISON DIVISION
ZERO BLEED

POWER	MACH	SHR	SFC	WF	FN	ESHP	WCIN	TNOZ	PNOZ	RC	NO
CLIMB	0.50	93	55	1007	33	77	33	369	55	66	66
W.C.	0.50	193	111	1007	77	33	77	369	11	44	033
CLIMB	0.50	33	33	893	55	33	33	369	55	44	033
W.C.	0.50	330	42	653	2	33	33	2265	66	44	033
CLIMB	0.50	57	42	209	2	33	33	105	55	44	033
W.C.	0.50	57	42	209	2	33	33	386	55	44	033
CLIMB	0.60	48	42	918	2	33	33	386	55	44	033
W.C.	0.60	48	42	918	2	33	33	222	55	44	033
CLIMB	0.60	48	42	32	2	33	33	1037	55	44	033
W.C.	0.60	48	42	32	2	33	33	409	55	44	033
CLIMB	0.70	23	42	201	2	33	33	409	55	44	033
W.C.	0.70	23	42	201	2	33	33	328	55	44	033
CLIMB	0.70	102	42	192	2	33	33	17	55	44	033
W.C.	0.70	102	42	192	2	33	33	108	55	44	033
CLIMB	0.70	102	42	507	2	33	33	53	55	44	033
W.C.	0.70	102	42	507	2	33	33	27	55	44	033
CLIMB	0.75	59	42	27	2	33	33	27	55	44	033
W.C.	0.75	59	42	27	2	33	33	27	55	44	033
CLIMB	0.75	59	42	27	2	33	33	27	55	44	033
W.C.	0.75	59	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2	33	33	27	55	44	033
CLIMB	0.80	10	42	27	2	33	33	27	55	44	033
W.C.	0.80	10	42	27	2</						

DETROIT DIESEL ALLISON DIVISION

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

0 FEET ALTITUDE

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DETROIT DIESEL ALLISON DIVISION

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GENERAL MOTORS CORPORATION

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

5000 FEET ALTITUDE

POWER	MACH	PER-PFI-INLET-BEC	DEC-SHP	PER-PCI-BLEED-DEL-FR	K	NO
CLIMB M.C.	0.20	110.2	155.5	4.6	7.7	0049
	0.20	102.5	155.5	3.0	7.7	0050
	0.20	102.5	155.5	3.0	7.7	0051
	0.20	102.5	155.5	3.0	7.7	0052
	0.20	102.5	155.5	3.0	7.7	0053
	0.20	102.5	155.5	3.0	7.7	0054
	0.20	102.5	155.5	3.0	7.7	0055
	0.20	102.5	155.5	3.0	7.7	0056
	0.20	102.5	155.5	3.0	7.7	0057
	0.20	102.5	155.5	3.0	7.7	0058
	0.20	102.5	155.5	3.0	7.7	0059
	0.20	102.5	155.5	3.0	7.7	0060
	0.20	102.5	155.5	3.0	7.7	0061
	0.20	102.5	155.5	3.0	7.7	0062
	0.20	102.5	155.5	3.0	7.7	0063
	0.20	102.5	155.5	3.0	7.7	0064
	0.20	102.5	155.5	3.0	7.7	0065
	0.20	102.5	155.5	3.0	7.7	0066
	0.20	102.5	155.5	3.0	7.7	0067
	0.20	102.5	155.5	3.0	7.7	0068
	0.20	102.5	155.5	3.0	7.7	0069
	0.20	102.5	155.5	3.0	7.7	0070
	0.20	102.5	155.5	3.0	7.7	0071
	0.20	102.5	155.5	3.0	7.7	0072
	0.20	102.5	155.5	3.0	7.7	0073
	0.20	102.5	155.5	3.0	7.7	0074
	0.20	102.5	155.5	3.0	7.7	0075
	0.20	102.5	155.5	3.0	7.7	0076
	0.20	102.5	155.5	3.0	7.7	0077
	0.20	102.5	155.5	3.0	7.7	0078
	0.20	102.5	155.5	3.0	7.7	0079
	0.20	102.5	155.5	3.0	7.7	0080
	0.20	102.5	155.5	3.0	7.7	0081
	0.20	102.5	155.5	3.0	7.7	0082
	0.20	102.5	155.5	3.0	7.7	0083
	0.20	102.5	155.5	3.0	7.7	0084
	0.20	102.5	155.5	3.0	7.7	0085
	0.20	102.5	155.5	3.0	7.7	0086
	0.20	102.5	155.5	3.0	7.7	0087
	0.20	102.5	155.5	3.0	7.7	0088

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GENERAL MOTORS CORPORATION

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS
10000 FEET ALTITUDE

POWER	MACH	PER-SCI-INLET-BE	DEL-SAP	PER-SCI-BLEED	DEL-FN	K	NO
CLIMB M.C.	0.00	97.18	5.0	6.0	15.0	60	0090
	0.02	97.10	5.0	6.0	15.0	60	0091
	0.04	97.05	5.0	6.0	15.0	60	0092
	0.06	97.00	5.0	6.0	15.0	60	0093
	0.08	96.95	5.0	6.0	15.0	60	0094
	0.10	96.90	5.0	6.0	15.0	60	0095
	0.12	96.85	5.0	6.0	15.0	60	0096
	0.14	96.80	5.0	6.0	15.0	60	0097
	0.16	96.75	5.0	6.0	15.0	60	0098
	0.18	96.70	5.0	6.0	15.0	60	0099
	0.20	96.65	5.0	6.0	15.0	60	0100
	0.22	96.60	5.0	6.0	15.0	60	0101
	0.24	96.55	5.0	6.0	15.0	60	0102
	0.26	96.50	5.0	6.0	15.0	60	0103
	0.28	96.45	5.0	6.0	15.0	60	0104
	0.30	96.40	5.0	6.0	15.0	60	0105
	0.32	96.35	5.0	6.0	15.0	60	0106
	0.34	96.30	5.0	6.0	15.0	60	0107
	0.36	96.25	5.0	6.0	15.0	60	0108
	0.38	96.20	5.0	6.0	15.0	60	0109
	0.40	96.15	5.0	6.0	15.0	60	0110
	0.42	96.10	5.0	6.0	15.0	60	0111
	0.44	96.05	5.0	6.0	15.0	60	0112
	0.46	96.00	5.0	6.0	15.0	60	0113
	0.48	95.95	5.0	6.0	15.0	60	0114
	0.50	95.90	5.0	6.0	15.0	60	0115
	0.52	95.85	5.0	6.0	15.0	60	0116
	0.54	95.80	5.0	6.0	15.0	60	0117
	0.56	95.75	5.0	6.0	15.0	60	0118
	0.58	95.70	5.0	6.0	15.0	60	0119
	0.60	95.65	5.0	6.0	15.0	60	0120
	0.62	95.60	5.0	6.0	15.0	60	0121
	0.64	95.55	5.0	6.0	15.0	60	0122
	0.66	95.50	5.0	6.0	15.0	60	0123
	0.68	95.45	5.0	6.0	15.0	60	0124
	0.70	95.40	5.0	6.0	15.0	60	0125
	0.72	95.35	5.0	6.0	15.0	60	0126
	0.74	95.30	5.0	6.0	15.0	60	0127
	0.76	95.25	5.0	6.0	15.0	60	0128
	0.78	95.20	5.0	6.0	15.0	60	0129
	0.80	95.15	5.0	6.0	15.0	60	0130
	0.82	95.10	5.0	6.0	15.0	60	0131
	0.84	95.05	5.0	6.0	15.0	60	0132
	0.86	95.00	5.0	6.0	15.0	60	0133
	0.88	94.95	5.0	6.0	15.0	60	0134
	0.90	94.90	5.0	6.0	15.0	60	0135
	0.92	94.85	5.0	6.0	15.0	60	0136
	0.94	94.80	5.0	6.0	15.0	60	0137
	0.96	94.75	5.0	6.0	15.0	60	0138
	0.98	94.70	5.0	6.0	15.0	60	0139
	1.00	94.65	5.0	6.0	15.0	60	0140
	1.02	94.60	5.0	6.0	15.0	60	0141
	1.04	94.55	5.0	6.0	15.0	60	0142
	1.06	94.50	5.0	6.0	15.0	60	0143
	1.08	94.45	5.0	6.0	15.0	60	0144
	1.10	94.40	5.0	6.0	15.0	60	0145
	1.12	94.35	5.0	6.0	15.0	60	0146
	1.14	94.30	5.0	6.0	15.0	60	0147
	1.16	94.25	5.0	6.0	15.0	60	0148
	1.18	94.20	5.0	6.0	15.0	60	0149
	1.20	94.15	5.0	6.0	15.0	60	0150
	1.22	94.10	5.0	6.0	15.0	60	0151
	1.24	94.05	5.0	6.0	15.0	60	0152
	1.26	94.00	5.0	6.0	15.0	60	0153
	1.28	93.95	5.0	6.0	15.0	60	0154
	1.30	93.90	5.0	6.0	15.0	60	0155
	1.32	93.85	5.0	6.0	15.0	60	0156
	1.34	93.80	5.0	6.0	15.0	60	0157
	1.36	93.75	5.0	6.0	15.0	60	0158
	1.38	93.70	5.0	6.0	15.0	60	0159
	1.40	93.65	5.0	6.0	15.0	60	0160
	1.42	93.60	5.0	6.0	15.0	60	0161
	1.44	93.55	5.0	6.0	15.0	60	0162
	1.46	93.50	5.0	6.0	15.0	60	0163
	1.48	93.45	5.0	6.0	15.0	60	0164
	1.50	93.40	5.0	6.0	15.0	60	0165
	1.52	93.35	5.0	6.0	15.0	60	0166
	1.54	93.30	5.0	6.0	15.0	60	0167
	1.56	93.25	5.0	6.0	15.0	60	0168
	1.58	93.20	5.0	6.0	15.0	60	0169
	1.60	93.15	5.0	6.0	15.0	60	0170
	1.62	93.10	5.0	6.0	15.0	60	0171
	1.64	93.05	5.0	6.0	15.0	60	0172
	1.66	93.00	5.0	6.0	15.0	60	0173
	1.68	92.95	5.0	6.0	15.0	60	0174
	1.70	92.90	5.0	6.0	15.0	60	0175
	1.72	92.85	5.0	6.0	15.0	60	0176
	1.74	92.80	5.0	6.0	15.0	60	0177
	1.76	92.75	5.0	6.0	15.0	60	0178
	1.78	92.70	5.0	6.0	15.0	60	0179
	1.80	92.65	5.0	6.0	15.0	60	0180
	1.82	92.60	5.0	6.0	15.0	60	0181
	1.84	92.55	5.0	6.0	15.0	60	0182
	1.86	92.50	5.0	6.0	15.0	60	0183
	1.88	92.45	5.0	6.0	15.0	60	0184
	1.90	92.40	5.0	6.0	15.0	60	0185
	1.92	92.35	5.0	6.0	15.0	60	0186
	1.94	92.30	5.0	6.0	15.0	60	0187
	1.96	92.25	5.0	6.0	15.0	60	0188
	1.98	92.20	5.0	6.0	15.0	60	0189
	2.00	92.15	5.0	6.0	15.0	60	0190
	2.02	92.10	5.0	6.0	15.0	60	0191
	2.04	92.05	5.0	6.0	15.0	60	0192
	2.06	92.00	5.0	6.0	15.0	60	0193
	2.08	91.95	5.0	6.0	15.0	60	0194
	2.10	91.90	5.0	6.0	15.0	60	0195
	2.12	91.85	5.0	6.0	15.0	60	0196
	2.14	91.80	5.0	6.0	15.0	60	0197
	2.16	91.75	5.0	6.0	15.0	60	0198
	2.18	91.70	5.0	6.0	15.0	60	0199
	2.20	91.65	5.0	6.0	15.0	60	0200

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GENERAL MOTORS CORPORATION

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

15000 FEET ALTITUDE

[illegible]

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PC370-42 TURBOPROP

GENERAL MOTORS CORPORATION

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

25000 FEET ALTITUDE

[illegible]

DETROIT DIESEL ALLISON DIVISION

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PD370-42 TURBOPROP

GENERAL MOTORS CORPORATION

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS
30000 FEET ALTITUDE

POWER	MACH	PER-SCA-INLET-BLEED	DEL-SAP-DEL-BLEED	DEL-FCN	K	NO
CLIMB M.C.	0.50	5.2	1.1	0.0	3.5	0297
	0.52	5.4	1.1	0.0	3.5	0298
	0.55	5.7	1.1	0.0	3.5	0299
	0.58	6.0	1.1	0.0	3.5	0300
	0.60	6.2	1.1	0.0	3.5	0301
	0.62	6.4	1.1	0.0	3.5	0302
	0.64	6.6	1.1	0.0	3.5	0303
	0.66	6.8	1.1	0.0	3.5	0304
	0.68	7.0	1.1	0.0	3.5	0305
	0.70	7.2	1.1	0.0	3.5	0306
	0.72	7.4	1.1	0.0	3.5	0307
	0.74	7.6	1.1	0.0	3.5	0308
	0.76	7.8	1.1	0.0	3.5	0309
	0.78	8.0	1.1	0.0	3.5	0310
	0.80	8.2	1.1	0.0	3.5	0311
	0.82	8.4	1.1	0.0	3.5	0312
	0.84	8.6	1.1	0.0	3.5	0313
	0.86	8.8	1.1	0.0	3.5	0314
	0.88	9.0	1.1	0.0	3.5	0315
	0.90	9.2	1.1	0.0	3.5	0316
	0.92	9.4	1.1	0.0	3.5	0317
	0.94	9.6	1.1	0.0	3.5	0318
	0.96	9.8	1.1	0.0	3.5	0319
	0.98	10.0	1.1	0.0	3.5	0320
	1.00	10.2	1.1	0.0	3.5	0321
	1.02	10.4	1.1	0.0	3.5	0322
	1.04	10.6	1.1	0.0	3.5	0323
	1.06	10.8	1.1	0.0	3.5	0324
	1.08	11.0	1.1	0.0	3.5	0325
	1.10	11.2	1.1	0.0	3.5	0326
	1.12	11.4	1.1	0.0	3.5	0327
	1.14	11.6	1.1	0.0	3.5	0328
	1.16	11.8	1.1	0.0	3.5	0329
	1.18	12.0	1.1	0.0	3.5	0330
	1.20	12.2	1.1	0.0	3.5	0331
	1.22	12.4	1.1	0.0	3.5	0332
	1.24	12.6	1.1	0.0	3.5	0333
	1.26	12.8	1.1	0.0	3.5	0334
	1.28	13.0	1.1	0.0	3.5	0335
	1.30	13.2	1.1	0.0	3.5	0336
	1.32	13.4	1.1	0.0	3.5	0337
	1.34	13.6	1.1	0.0	3.5	0338
	1.36	13.8	1.1	0.0	3.5	0339
	1.38	14.0	1.1	0.0	3.5	0340
	1.40	14.2	1.1	0.0	3.5	0341
	1.42	14.4	1.1	0.0	3.5	0342
	1.44	14.6	1.1	0.0	3.5	0343
	1.46	14.8	1.1	0.0	3.5	0344
	1.48	15.0	1.1	0.0	3.5	0345
	1.50	15.2	1.1	0.0	3.5	0346
	1.52	15.4	1.1	0.0	3.5	0347
	1.54	15.6	1.1	0.0	3.5	0348
	1.56	15.8	1.1	0.0	3.5	0349
	1.58	16.0	1.1	0.0	3.5	0350
	1.60	16.2	1.1	0.0	3.5	0351
	1.62	16.4	1.1	0.0	3.5	0352
	1.64	16.6	1.1	0.0	3.5	0353
	1.66	16.8	1.1	0.0	3.5	0354
	1.68	17.0	1.1	0.0	3.5	0355
	1.70	17.2	1.1	0.0	3.5	0356
	1.72	17.4	1.1	0.0	3.5	0357
	1.74	17.6	1.1	0.0	3.5	0358
	1.76	17.8	1.1	0.0	3.5	0359
	1.78	18.0	1.1	0.0	3.5	0360
	1.80	18.2	1.1	0.0	3.5	0361
	1.82	18.4	1.1	0.0	3.5	0362
	1.84	18.6	1.1	0.0	3.5	0363
	1.86	18.8	1.1	0.0	3.5	0364
	1.88	19.0	1.1	0.0	3.5	0365
	1.90	19.2	1.1	0.0	3.5	0366
	1.92	19.4	1.1	0.0	3.5	0367
	1.94	19.6	1.1	0.0	3.5	0368
	1.96	19.8	1.1	0.0	3.5	0369
	1.98	20.0	1.1	0.0	3.5	0370
	2.00	20.2	1.1	0.0	3.5	0371
	2.02	20.4	1.1	0.0	3.5	0372
	2.04	20.6	1.1	0.0	3.5	0373
	2.06	20.8	1.1	0.0	3.5	0374
	2.08	21.0	1.1	0.0	3.5	0375
	2.10	21.2	1.1	0.0	3.5	0376
	2.12	21.4	1.1	0.0	3.5	0377
	2.14	21.6	1.1	0.0	3.5	0378
	2.16	21.8	1.1	0.0	3.5	0379
	2.18	22.0	1.1	0.0	3.5	0380
	2.20	22.2	1.1	0.0	3.5	0381
	2.22	22.4	1.1	0.0	3.5	0382
	2.24	22.6	1.1	0.0	3.5	0383
	2.26	22.8	1.1	0.0	3.5	0384
	2.28	23.0	1.1	0.0	3.5	0385
	2.30	23.2	1.1	0.0	3.5	0386
	2.32	23.4	1.1	0.0	3.5	0387
	2.34	23.6	1.1	0.0	3.5	0388
	2.36	23.8	1.1	0.0	3.5	0389
	2.38	24.0	1.1	0.0	3.5	0390
	2.40	24.2	1.1	0.0	3.5	0391
	2.42	24.4	1.1	0.0	3.5	0392
	2.44	24.6	1.1	0.0	3.5	0393
	2.46	24.8	1.1	0.0	3.5	0394
	2.48	25.0	1.1	0.0	3.5	0395
	2.50	25.2	1.1	0.0	3.5	0396
	2.52	25.4	1.1	0.0	3.5	0397
	2.54	25.6	1.1	0.0	3.5	0398
	2.56	25.8	1.1	0.0	3.5	0399
	2.58	26.0	1.1	0.0	3.5	0400
	2.60	26.2	1.1	0.0	3.5	0401
	2.62	26.4	1.1	0.0	3.5	0402
	2.64	26.6	1.1	0.0	3.5	0403
	2.66	26.8	1.1	0.0	3.5	0404
	2.68	27.0	1.1	0.0	3.5	0405
	2.70	27.2	1.1	0.0	3.5	0406
	2.72	27.4	1.1	0.0	3.5	0407
	2.74	27.6	1.1	0.0	3.5	0408
	2.76	27.8	1.1	0.0	3.5	0409
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	2.98	30.0	1.1	0.0	3.5	0420
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	3.02	30.4	1.1	0.0	3.5	0422
	3.04	30.6	1.1	0.0	3.5	0423
	3.06	30.8	1.1	0.0	3.5	0424
	3.08	31.0	1.1	0.0	3.5	0425
	3.10	31.2	1.1	0.0	3.5	0426
	3.12	31.4	1.1	0.0	3.5	0427
	3.14	31.6	1.1	0.0	3.5	0428
	3.16	31.8	1.1	0.0	3.5	0429
	3.18	32.0	1.1	0.0	3.5	0430
	3.20	32.2	1.1	0.0	3.5	0431
	3.22	32.4	1.1	0.0	3.5	0432
	3.24	32.6	1.1	0.0	3.5	0433
	3.26	32.8	1.1	0.0	3.5	0434
	3.28	33.0	1.1	0.0	3.5	0435
	3.30	33.2	1.1	0.0	3.5	0436
	3.32	33.4	1.1	0.0	3.5	0437
	3.34	33.6	1.1	0.0	3.5	0438
	3.36	33.8	1.1	0.0	3.5	0439
	3.38	34.0	1.1	0.0	3.5	0440
	3.40	34.2	1.1	0.0	3.5	0441
	3.42	34.4	1.1	0.0	3.5	0442
	3.44	34.6	1.1	0.0	3.5	0443
	3.46	34.8	1.1	0.0	3.5	0444
	3.48	35.0	1.1	0.0	3.5	0445
	3.50	35.2	1.1	0.0	3.5	0446
	3.52	35.4	1.1	0.0	3.5	0447
	3.54	35.6	1.1	0.0	3.5	0448
	3.56	35.8	1.1	0.0	3.5	0449
	3.58	36.0	1.1	0.0	3.5	0450
	3.60	36.2	1.1	0.0	3.5	0451
	3.62	36.4	1.1	0.0	3.5	0452
	3.64	36.6	1.1	0.0	3.5	0453
	3.66	36.8	1.1	0.0	3.5	0454
	3.68	37.0	1.1	0.0	3.5	0455
	3.70	37.2	1.1	0.0	3.5	0456
	3.72	37.4	1.1	0.0	3.5	0457
	3.74	37.6	1.1	0.0	3.5	0458
	3.76	37.8	1.1	0.0	3.5	0459
	3.78	38.0	1.1	0.0	3.5	0460
	3.80	38.2	1.1	0.0	3.5	0461
	3.82	38.4	1.1	0.0	3.5	0462
	3.84	38.6	1.1	0.0	3.5	0463
	3.86	38.8	1.1	0.0	3.5	0464
	3.88	39.0	1.1	0.0	3.5	0465
	3.90	39.2	1.1	0.0	3.5	0466
	3.92	39.4	1.1	0.0	3.5	0467
	3.94	39.6	1.1	0.0	3.5	0468
	3.96	39.8	1.1	0.0	3.5	0469
	3.98	40.0	1.1	0.0	3.5	0470
	4.00	40.2	1.1	0.0	3.5	0471
	4.02	40.4	1.1	0.0	3.5	0472
	4.04	40.6	1.1	0.0	3.5	0473
	4.06	40.8	1.1	0.0	3.5	0474
	4.08	41.0	1.1	0.0	3.5	0475
	4.10	41.2	1.1	0.0	3.5	0476
	4.12	41.4	1.1	0.0	3.5	0477
	4.14	41.6	1.1	0.0	3.5	0478
	4.16	41.8	1.1	0.0	3.5	0479
	4.18	42.0	1.1	0.0	3.5	0480
	4.20	42.2	1.1	0.0	3.5	0481
	4.22	42.4	1.1	0.0	3.5	0482
	4.24	42.6	1.1	0.0	3.5	0483
	4.26	42.8	1.1	0.0	3.5	0484
	4.28	43.0	1.1	0.0	3.5	0485
	4.30	43.2	1.1	0.0	3.5	0486
	4.32	43.4	1.1	0.0	3.5	0487
	4.34	43.6	1.1	0.0	3.5	0488
	4.36	43.8	1.1	0.0	3.5	0489
	4.38	44.0	1.1	0.0	3.5	0490
	4.40	44.2	1.1	0.0	3.5	0491
	4.42	44.4	1.1	0.0	3.5	0492
	4.44	44.6	1.1	0.0	3.5	0493
	4.46	44.8	1.1	0.0	3.5	0494
	4.48	45.0	1.1	0.0	3.5	0495
	4.50	45.2	1.1	0.0	3.5	0496
	4.52	45.4	1.1	0.0	3.5	0497
	4.54	45.6	1.1	0.0	3.5	0498
	4.56	45.8	1.1	0.0	3.5	0499
	4.58	46.0	1.1	0.0	3.5	0500
	4.60	46.2	1.1	0.0	3.	

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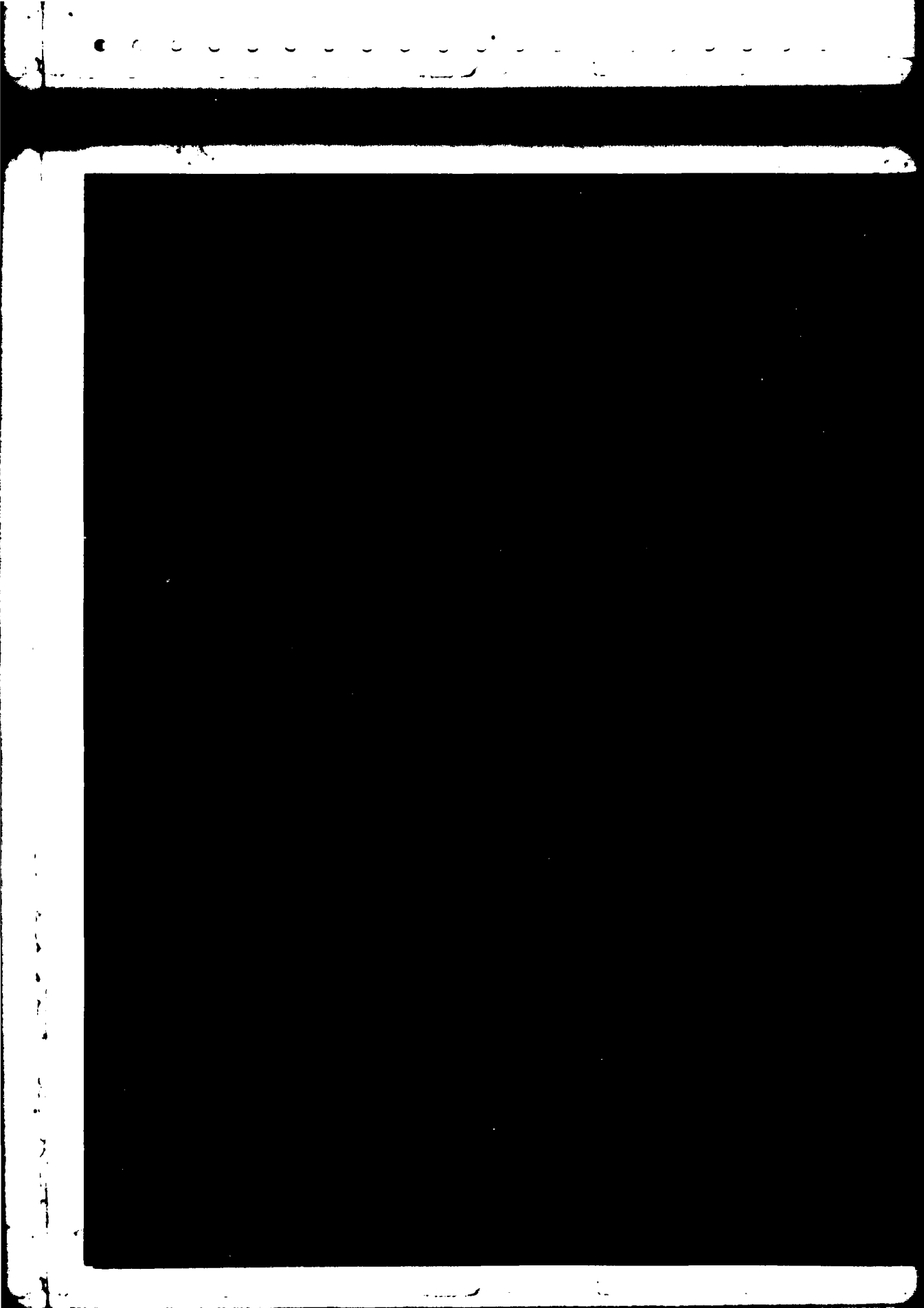
PD370-42 TURBOPROP

GENERAL MOTORS CORPORATION

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

45000 FEET ALTITUDE

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